

In the Claims:

1. (Original) An arrangement for detecting a shaft break on a rotor of a first turbine (10), particularly a medium pressure turbine of a gas turbine, particularly of an aircraft engine, whereby a second turbine (11), particularly a low pressure turbine, is positioned downstream of the first turbine (10), with an operator element (16) positioned between the rotor of the first turbine (10) and a stator of the second turbine (11) radially inwardly relative to a flow channel, and with a sensor element (21) guided in the stator of the second turbine (11), in order to convert a shaft break, detected by the radially inwardly positioned operator element (16), into an electrical signal and to transmit this electrical signal to a switching element which is positioned radially outwardly relative to the flow channel on a housing of the gas turbine.

2. (Original) The arrangement of claim 1, characterized in that the operator element (16) is positioned between a last rotor blade ring of the first turbine (10), as seen in the flow direction, and a first guide vane ring of the second turbine (11), as seen in the flow direction.

1 3. (Original) The arrangement of claim 2, characterized in
2 that the operator element (16) is positioned radially
3 inwardly and neighboring to a rotor disk (12) of the last
4 rotor blade ring, as seen in the flow direction, of the
5 first turbine (10).

Claims 4 to 8 (canceled).

1 9. (Original) A gas turbine, particularly an aircraft engine,
2 with at least two compressors, at least one combustion
3 chamber, and at least two turbines, and with an arrangement
4 for detecting a shaft break in a rotor of a first turbine
5 (10), particularly a medium pressure turbine, whereby a
6 second turbine (11), particularly a low pressure turbine,
7 is positioned downstream of the first turbine,
8 characterized in that an operator element (16) is
9 positioned between the rotor of the first turbine (10) and
10 a stator of the second turbine (11) radially inwardly
11 relative to a flow channel, and in that a sensor element
12 (21) is guided in the stator of the second turbine (11) in
13 order to convert a shaft break detected by the radially
14 inwardly positioned operator element (16) into an
15 electrical signal and to transmit this electrical signal to
16 a switching element which is positioned radially outwardly
17 relative to the flow channel on a housing of the gas
18 turbine.

1 10. (Original) The gas turbine of claim 9, characterized in
2 that the operator element (16) is positioned between a last
3 rotor blade ring of the first turbine (10), as seen in the
4 flow direction, and a first guide vane ring of the second
5 turbine (11), as seen in the flow direction.

1 11. (Original) The gas turbine of claim 10, characterized in
2 that the operator element (16) is positioned radially
3 inwardly and neighboring to a rotor disk (12) of the last
4 rotor blade ring, as seen in the flow direction, of the
5 first turbine (10).

Claims 12 to 16 (Canceled).

[REMARKS FOLLOW ON NEXT PAGE]